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Computational dynamics of reduced coupled multibodyfluid system in Lie group setting

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Motivation

Why geometric approach ?

- Valuable mathematical insight to various problems of mechanics and engineering
 - traditional link between analytical mechanics and differential-geometry
 - modelling of finite (3D) rotations
 - numerical treatment of configuration constraints
 - \Rightarrow call 'naturally' for geometric mathematical framework
- Design of structure-preserving numerical integration methods
 - a) 'Smart' integration procedures that respect
 - underlying kinematic and dynamic structure of the system \rightarrow qualitative behaviour
 - integration methods on manifolds and Lie-groups
- Non-linear control design
- Connections to other fields of physics and engineering
 - mechanics of continuous media /multi-physics
 - \rightarrow fluid mechanics

/ fluid-structure interaction

- \rightarrow magnetism, optics
- \rightarrow nano-scale systems





Motivation / Fluid-structure interaction

- discretisation of fluid domain + coupling with solid motion
- Ioosely coupled algorithms / ALE formulation
- moving FV mesh: problems with accuracy, stability...



meshless methods: SPH, Lattice-Boltzmann Method ...



- configuration space: Lie groups
 - ▶ rigid body ⇒ SE(3) (SO(3)× \Re^3); SE(2)
 - ➢ incompressible fluid ⇒ $Diff_{Vol}(\mathcal{F})$

Arnold, V.: Sur la géométrie différentielle des groupes de Lie de dimension infinie et ses applications à l'hydrodynamique des fluides parfaits. Annales de l'Institut Fourier 16(1), 319-361 (1966) ${\mathcal F}$: domain in ${\mathcal R}^2$ or ${\mathcal R}^3$

(infinite - dimensional group)





Incompressible ideal fluid







Lie algebra and dual spaces

Velocities

- at the identity: Lie algebras rigid body \Rightarrow se(3) (so(3)× \Re^3); se(2) $\widetilde{\omega} \in$ so(3)
- incompressible fluid $\Rightarrow V_{Div}(\mathscr{F}_0)$ divergence-free vector fields in \mathscr{F}_0 tangent to the boundary $\partial \mathscr{F}_0$
- Dual spaces of Lie algebras
 - ► rigid body $\Rightarrow y = I\omega \in so(3)^*$... angular momentum
 - ➢ incompresible fluid ⇒ vorticity field (circulation) V^{*}_{Div}(𝔅) for a flow v ∈ V_{Div}(𝔅)



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Dynamics

- Rotation motion of rigid body
 - geodesic on G = SO(3) equipped with left-invariant
 Riemannian metric

Motion of ideal incompressible fluid

- Seodesic on $G = \text{Diff}_{Vol}(\mathcal{F})$ with right-invariant Riemannian metric
- Dynamics evolves on:

TG: 'positions' + velocities (*Lie algebra*): body, fluid T^*G : 'positions' + momentum (body) / vorticity (fluid) (dual Lie algebra)





Symmetries

Rigid body G = SO(3) $L_h: G \to G; \ L_h(g): hg$ $G = \mathbf{R} \in SO(3)$ $KE_{body} = \frac{1}{2}\omega^T I\omega$

Incompressible fluid $G = \text{Diff}_{\text{Vol}}(\mathscr{F})$ $R_h: G \to G; \quad R_h(g): gh$



$$KE_{fluid} = \frac{1}{2} \iint_{\mathcal{F}} \mathbf{v}^2 dv$$

 $L_h, R_h \rightarrow KE_{invariant} \xrightarrow{Noether theorem} momentum map J conserved$





Momentum map: *SO*(3)

• Momentum map associated with G = SO(3)

$$J_G: T^*G \to so(3)^*$$

► angular momentum $y = I\omega \in so(3)^*$ conserved ⇒ 'coadjoint orbits' preserved



Euler equation : $\frac{dy}{dt} = -\omega \times y$ - solution in the form: $y^{n+1} = \mathbf{R}^{\mathrm{T}}(t) y^{n}$, $y^{n+1} = \mathrm{Ad}_{G}^{*} y^{n}$ $G = \mathbf{R} \in SO(3)$

Forward Euler on Lie-group

KE and Lagrangian: G - invariant



Momentum map: $\text{Diff}_{Vol}(\mathcal{F})$

• Momentum map associated with $\mathrm{Diff}_{\mathrm{Vol}}(\mathcal{F})$ $J_{\text{Diff}_{\text{Vol}}}: T^*\text{Diff}_{\text{Vol}} \to V_{\text{Div}}^*(\mathcal{F}_{o})$ vorticity advection / $V_{\text{Div}}^*(\mathcal{F}_0)$: vorticity + circulation $[\alpha] \in V_{\text{Div}}^*(\mathscr{F}_{o}); [\alpha] \mapsto (\mathbf{d}\alpha, \Gamma \coloneqq \int \alpha)$ fluid *KE* and Lagrangian \Rightarrow Diff_{Vol} - invariant momentum map is conserved particle relabeling symmetry Kelvin's theorem: circulation around closed curve -> preserved; vorticity is advected coadjoint orbits preserved !!





Euler equations: body rotation / fluid dynamics

Second Euler teorem on arbitrary Lie group G (left invariant metric):

 $\frac{d\mu}{dt} = ad_{\omega_{Lie}}^* \mu \ ; \ \mu \in g^*$ rigid body: $\mu = y \in so(3)^*$ $\dot{y} = ad_{\omega}^* y = -\omega \times y \implies y = Ad_G^* y$ ideal fluid: $[\alpha] \in V_{Div}^*(\mathscr{F}_0); \ [\alpha] \mapsto (\mathbf{d}\alpha, \Gamma)$



 $(\text{in } \mathcal{R}^3)... \quad \frac{\partial \mathbf{v}}{\partial t} = v \times \text{curl} v - \text{grad} p$

 $\dot{v} = -\mathbf{B}(\mathbf{v}, \mathbf{v})$; $\mathbf{B}(c, a) = \operatorname{curl} c \times a + \operatorname{grad} p$

- Solution: preservation of coadjoint orbits
 - body angular momentum; fluid circulation: $\operatorname{Ad}^*_{\operatorname{Diff}_{\operatorname{Vol}}}(\mathbf{d}\alpha,\Gamma) = (\mathbf{d}(\operatorname{Diff}_{\operatorname{Vol}}\alpha),\Gamma)$





Monolithic coupling

Geometric Fluid-MBS monolithic coupling

 $\mathcal{M} = \mathcal{B}_1 \cup \cdots \cup \mathcal{B}_k \cup \mathcal{F}$

Assumptions:

✓ potential flow

v no vorticity (no circulation!)

✓ *F* is connected

> fluid velocity field: $\mathbf{v} = \nabla \phi$

: incompresibility $\Rightarrow \Delta \phi = 0$ in \mathcal{F}

Configuration space: $Q(\mathcal{M}) := \{ f : \mathcal{M} \to \mathcal{M} \}$

 $q_{\mathcal{F}} \in \operatorname{Diff}_{\operatorname{Vol}}(\mathcal{F})$ $q_{\beta_i} \sim \operatorname{rigid} \operatorname{body} \operatorname{motion} \beta_i$ $G_{body} \subseteq SE(3) \times \ldots \times SE(3) / \mathcal{R}^3 \times SO(3) \times \ldots \times \mathcal{R}^3 \times SO(3)$

OF AERONAUTICAL ENGINEERING CHAIR OF FLIGHT VEHICLE DYNAMICS Fluid-Body model reduction / particle symmetr $G = \mathbf{R} \in SO(3)$ **Rigid body** $L(\mathbf{R}, \dot{\mathbf{R}}) = tr((\mathbf{R}^{-1}\dot{\mathbf{R}})^T I \mathbf{R}^{-1}\dot{\mathbf{R}}) \rightarrow \frac{1}{2}\omega^T I\omega$ $SO(3) \xrightarrow{Lie-Poisson} (Euler-Poincare)$ **Euler equations** (I)in Lie-Poisson form: $\dot{y} = -\omega \times y$, $y = I\omega \in so(3)^*$ $(I\dot{\omega} = I\omega \times \omega)$ (*no* **R** !!) **Rigid body + ideal fluid / particle relabeling symmetry** $\frac{Symplectic (\Gamma=0)}{(right)} \quad \text{Diff}_{Vol}(\mathcal{F})$ $\left[(q_{\beta_i}, q_{\mathcal{F}}, y_{\beta_i}, \alpha) \right] \mapsto (q_{\beta_i}, m(y_{\beta_i}))$ $T^* O \rightarrow T^* SE(3)$ (no fluid variables!!)



Fluid-MBS model reduction / particle symmetry

- **Reduction by the fluid** Diff_{vol} symmetry:
 - particle relabeling symmetry $\mathrm{Diff}_{\mathrm{vol}}(\mathcal{F})$
 - Fluid: Kelvin's theorem -> circulation preserved
 - \succ bodies: do not affect dynamics of \mathscr{B}_i

.... reduction at zero circulation:

 $(T^*Q)_{\Gamma=0} = \mathcal{J}^{-1}(O) / \operatorname{Diff}_{\operatorname{vol}}(\mathcal{F})$

▶ elimination of fluid variables $T^*Q \rightarrow T^*G_{body}$

reduced F-MBS configuration space: configuration space G_{body} of the submerged solids

 \succ reduced Lagrangian of F-MBS: function on TG_{body}

 \succ effect of fluid: added inertias to \mathscr{B}_i





Added inertias

Lagrangian F-MBS: $KE = KE_{body} + KE_{fluid}$

> KE_{fluid} : \mathcal{B}_i added inertias ; fluid velocity $f(\mathbf{v}_i \text{ and } \omega_i)$ velocity -> determined from velocity potential ϕ

$$\Delta \phi \,{=}\, 0\,$$
 in ${oldsymbol{\mathscr{F}}}$

- with prescribed velocities at $\partial \mathscr{B}_i$

Neumann problem for Laplace equation (linear !) with boundary conditions:

$$\nabla \boldsymbol{\phi} \cdot \mathbf{n}_i = \mathbf{v}_i \cdot \mathbf{n}_i \quad \text{in } \partial \boldsymbol{\mathscr{B}}_i$$
$$\nabla \boldsymbol{\phi} = 0 \qquad \text{at } \infty$$

- solved for translational and rotational potentials φ_i and χ_i after transformation (by superposition) into the form:

$$\phi = \sum_{i=1}^{N} (\omega_i \cdot \boldsymbol{\chi}_i + \mathbf{v}_i \cdot \boldsymbol{\varphi}_i)$$

- replaced by boundary value problem -> solved via BEM





Numerical efficiency

Fluid volume discretization \Rightarrow **Boundary** surface discretization

Multiple orders of magnitude fewer variables







MBS configuration and state space

MBS state space:

$$S = \mathcal{R}^3 \times SO(3) \times \ldots \times \mathcal{R}^3 \times so(3) \cong TG_{body}$$

with elements:

$$x = (\mathbf{r}_1, \mathbf{R}_1, \dots, \mathbf{r}_k, \mathbf{R}_k, \mathbf{v}_1, \widetilde{\boldsymbol{\omega}}_1, \dots, \mathbf{v}_k, \widetilde{\boldsymbol{\omega}}_k)$$

Lie group with Lie algebra with the element

 $z = (\mathbf{v}_1, \widetilde{\boldsymbol{\omega}}_1, \dots, \mathbf{v}_k, \widetilde{\boldsymbol{\omega}}_k, \dot{\mathbf{v}}_1, \dot{\boldsymbol{\omega}}_1, \dots, \dot{\mathbf{v}}_k, \dot{\boldsymbol{\omega}}_k)$

Lagrangian of F-MBS: function of x only $\mathcal{M} = \mathcal{B}_1 \cup \ldots \cup \mathcal{B}_k \cup \mathcal{F}$





Numerical integration: F-MBS system

 $\Re^3 \times SO(3)$

• Numerical integration ... $M \sim added mass$; $Q=Q_{ext}+Q_{ideal}+Q_{vort}$



- Munthe-Kaas type of integration algorithm:
- RK scheme applied to substitution ODE in Lie-algebra S
 - ... extended to DAE index 1 on Lie-groups





Results (MBS + fluid flow without circulation)

Bodies with blunt edges, no circulation:







Results (MBS + fluid flow without circulation)

Bodies with blunt edges, no circulation:







Results (MBS + fluid flow without circulation)

Bodies with blunt edges, no circulation:







Results (MBS + fluid flow without circulation)

Bodies with blunt edges, no circulation:





Fluid-Body / fluid flow with circulation

• Constant circulation Γ is preserved:

 $\mathrm{Ad}^*_{\mathrm{Diff}_{\mathrm{Vol}}}(0,\Gamma) = (0,\Gamma)$

Marsden, J.E. et al.: *Hamiltonian Reduction by Stages.* Lecture Notes in Math., vol. 1913, Springer-Verlag Berlin Heidelberg (2007)

(Kelvin's theorem) ... particle relabeling symetry

 Fluid-body symplectic reduced space: (T^{*}Q)_Γ = J⁻¹(Γ)/Diff_{vol}(𝔅)

 Isomorphism between (T^{*}Q)_Γ and T^{*}SE(2) [(q_{βi},q_𝔅,y_{βi},α)] → (q_{βi},m(y_{βi}))

Fluid variables (q_𝔅, α) eliminated
> added mass effect encoded by m, y_{𝔅i} ∈ se(2)*
T*Q → T*SE(2) → se*(2) > two stages reduction
~ Diff_{vol} action Q → principal bundle over SE(2) -:: NO
~ Neumann connection A: TQ → V_{Div} (𝔅)

-:: NC curvature: generates Kutta-Zhukowski force





Vorticity effects? ... Oscillating airfoil

- airfoil oscillating with frequency of 100 Hz
- simulated in *foam-extend* by using RANS with k-ω turbulence model
- effect of shedding trailing edge vortices is obvious
 - vorticity effects will be accounted for by enforcing a Kutta condition







Kutta condition

- Fluid is considered as ideal and vorticity effects are accounted for by enforcing a Kutta condition
- Kutta condition: velocity at the sharp edge must have meaningful physical value



Stagnation point: velocity can be zero or infinity







Enforcing Kutta condition









Vortex mechanism (UVLM)



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Enforcing Kutta condition

- Kutta condition: to be enforced by shedding vortices of appropriate "strength" from the trailing edge
- Procedure: distance from trailing edge at which to create a new vortex at each time step?
- P. Tallapragada, S. Kelly [2010, 2013, 2015, 2016], Hailong X., Ph.D. thesis, 2007, Mason R.J., Ph.D. thesis, CALTECH.











Flapping wing application

- Flapping wing of insect type
- large amplitudes of wing motion + complicated kinematics (...with high frequencies)
- very high accelerations \implies considerable 'added masses' effect
- highly unsteady flow (Strouhal number St ~ 1) / vortices at both edges !!
- low Reynolds number (*Re* = O (10² ~ 10⁰)) / but very thin boundary layer !!





Flapping wing: vortices and unsteady effects



- Ellington, C.P., van den Berg, C.,Willmott, A.P., Thomas, A.L.R.: *Leading-edge vortices in insect flight*. Nature 384, 626-630 (1996)
- Karásek, M., Muijres, F.T., Wagter, C.D., Remes, B.D.W., de Croon, G.C.H.E. *A tailless aerial robotic flapper reveals that flies use torque coupling in rapid banked turns*. Science 361, 1089–1094 (2018)

- LEV Leading edge vortex
- TEV1 Starting trailing edge vortex
- TEV2 Stopping trailing edge vortex
- TV1 Upper tip vortex
- TV2 Lower tip vortex





Flapping wing: quasi-steady aerodynamic model

: Lift enhancement due to the stabilized LEV during translation
: Reduction of the effective angle of attack due to the 'downwash'
: Lift enhancement due to the rapid rotation at the end of 'translation'
: 'Added mass effect'

Translational forces

$$F_{TL} = \int_0^R \frac{1}{2} \rho r^2 \dot{\phi}^2 c(r) C_{TL}(\alpha) dr$$
$$F_{TD} = \int_0^R \frac{1}{2} \rho r^2 \dot{\phi}^2 c(r) C_{TD}(\alpha) dr$$

Rotational force $F_{R} = \rho C_{R} R^{2} \dot{\phi} \dot{\eta} \overline{c}^{2} \int_{0}^{1} \hat{r} \hat{c}^{2}(\hat{r}) dr$

Added mass force



Lat. Drosophila melanogaster

Sane, S.P., Dickinson, M.H.: *The aerodynamic effects of wing rotation and a revised quasi-steady model of flapping flight.* Journal of Experimental Biology 205(8), 1087–1096 (2002)

$$F_A = \frac{\pi}{4} \rho R^2 \overline{c}^2 (\ddot{\phi} \sin \alpha + \dot{\phi} \dot{\alpha} \cos \alpha) \int_0^1 \hat{r} \hat{c}^2(\hat{r}) dr - \frac{\pi}{16} \rho \ddot{\alpha} R \overline{c}^3 \int_0^1 \hat{c}^2(\hat{r}) dr$$



DMOC optimisation procedure



$$\begin{split} \min_{q_d, u_d} J_d \left(q_d, u_d \right) &= \sum_{k=0}^{N-1} C \left(q_k, q_{k+1}, u_k \right) \\ q_0 &= q^0 \\ q_N &= q^T \\ p^0 + D_1 L_d \left(q_0, q_1 \right) + f_0^- &= 0 \\ -p^T + D_2 L_d \left(q_{N-1}, q_N \right) + f_{N-1}^+ &= 0 \\ d \left(q_{k-1}, q_k \right) + D_1 L_d \left(q_k, q_{k+1} \right) + f_{k-1}^+ + f_k^- &= 0 \\ h_d \left(q_k, q_{k+1}, u_k \right) &\geq 0 \end{split}$$

• Marsden, J.E., West, M.: *Discrete mechanics and variational integrators.* Acta Numerica 10, 357–514 (2001)

 Ober-Blöbaum, S., Junge, O., Marsden, J.E.: Discrete mechanics and optimal control: An analysis. ESAIM: Control, Optimisation and Calculus of Variations 17(2), 322–352 (2011)





Flapping wing: optimised flapping pattern





Mean initial power $\rightarrow 1.2144 \times 10^{-5}$ W Mean optimal power $\rightarrow 1.0907 \times 10^{-5}$ W Improvement $\rightarrow 10.2\%$





Flapping wing on Mars

	Earth	Mars
Atmospheric density [kg/m ³]	1.225	1.55×10^{-3}
Gravitational acceleration $\left[m/s^2 \right]$	9.81	3.72
Dynamic viscosity [kg/ms]	1.8×10^{-5}	$1.5 imes 10^{-5}$
Speed of sound [<i>m</i> /s]	343	247

• Mars aircraft fly at low Re numbers due to a significant difference in atmospheric density and only a slight difference in gravitational acceleration, dynamic viscosity and speed of sound.



Flapping wing on Mars: optimisation of hover







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